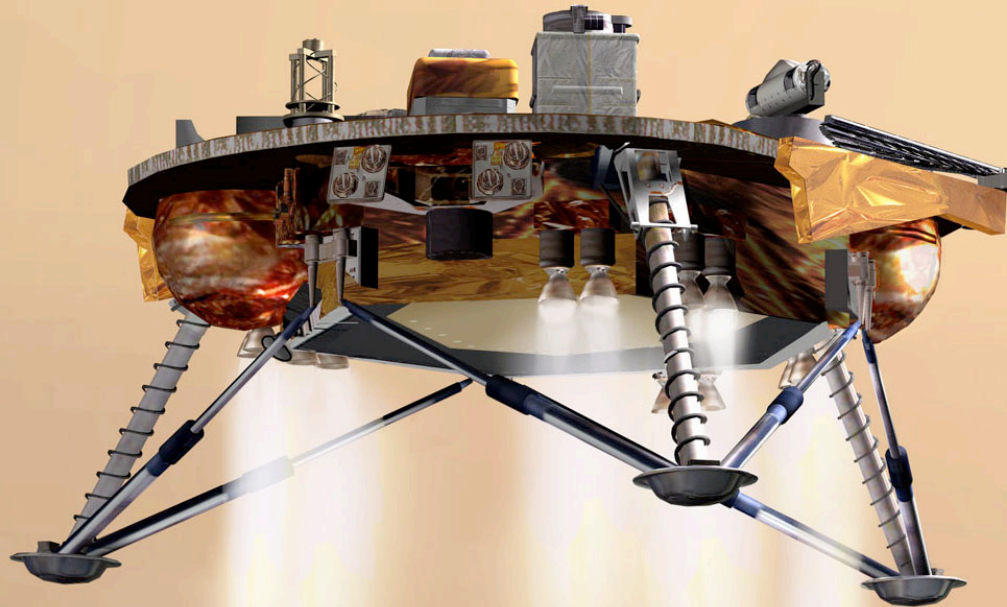
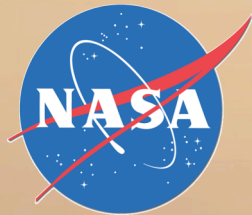


# Evolution of the Phoenix EDL System Architecture



**R. Grover, Jet Propulsion Laboratory  
P. Desai, NASA Langley Research Center**

**International Planetary Probe Workshop 5  
26 June 2007  
Bordeaux, France**

**National Aeronautics and Space Administration  
Jet Propulsion Laboratory California Institute of Technology**



National Aeronautics and Space  
Administration  
Jet Propulsion Laboratory  
California Institute of Technology

# Presentation Overview



- **The Phoenix Story**
- **Spacecraft Overview**
- **Phoenix EDL Overview**
- **Mission Design Comparison**
- **Hypersonic Subphase Evolution**
- **Terminal Descent Subphase Evolution**
- **Summary**



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# The Phoenix Story

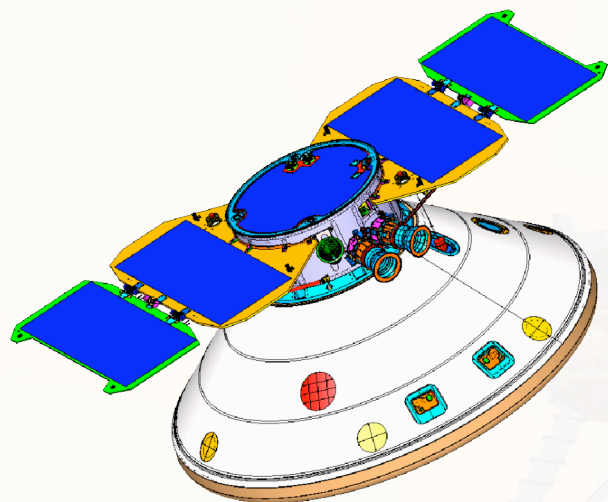


- **Started as Mars Surveyor 2001 Lander**
  - **Faster, better, cheaper spacecraft**
  - **Sister spacecraft of Mars Polar Lander**
  - **Cancelled after Mars Polar Lander failure in 1999**
    - **Not enough time to address findings of MPL failure review prior to 2001 launch window**
- **Reborn as Phoenix in 2003**
  - **Same spacecraft, modified science payloads**
  - **Enhanced radar**
  - **Addition of EDL communication subsystem**
  - **Enhanced test program**

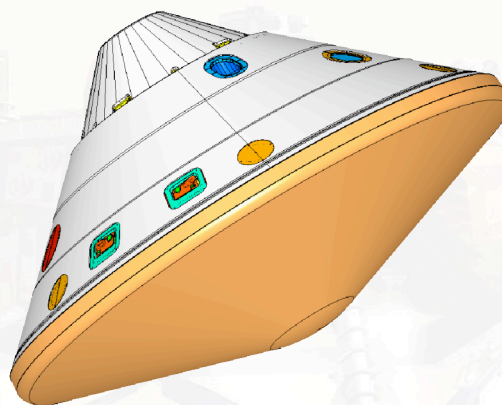


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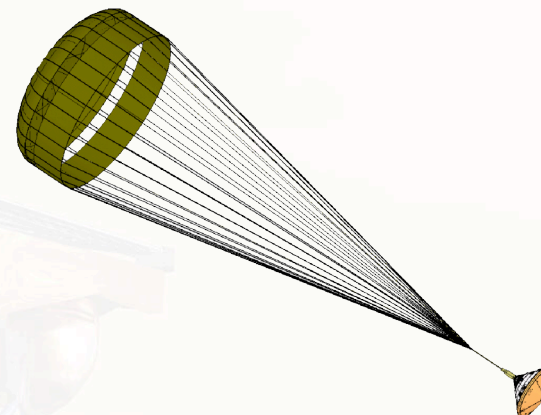
# Spacecraft Overview



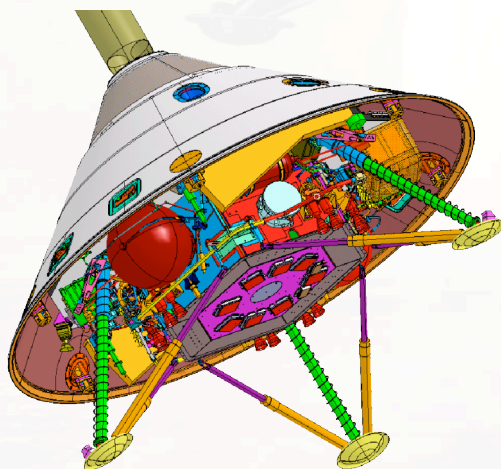
**Pre-Entry Configuration**



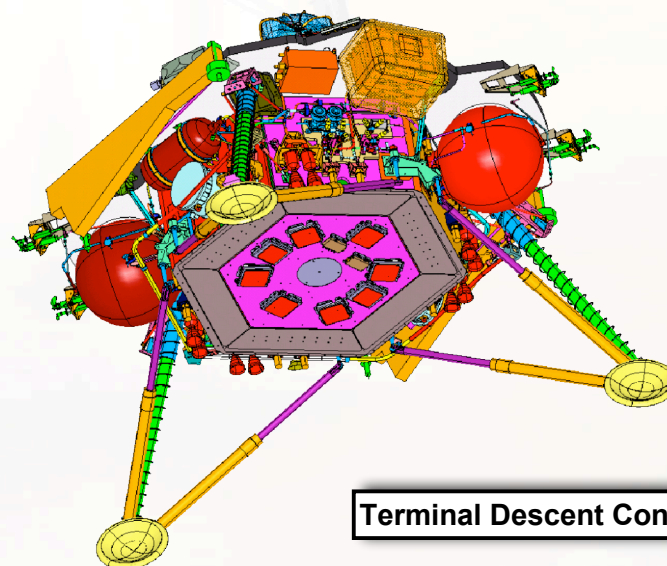
**Entry Configuration**



**Parachute Configuration**



**Post HS & Leg Deploy Configuration**

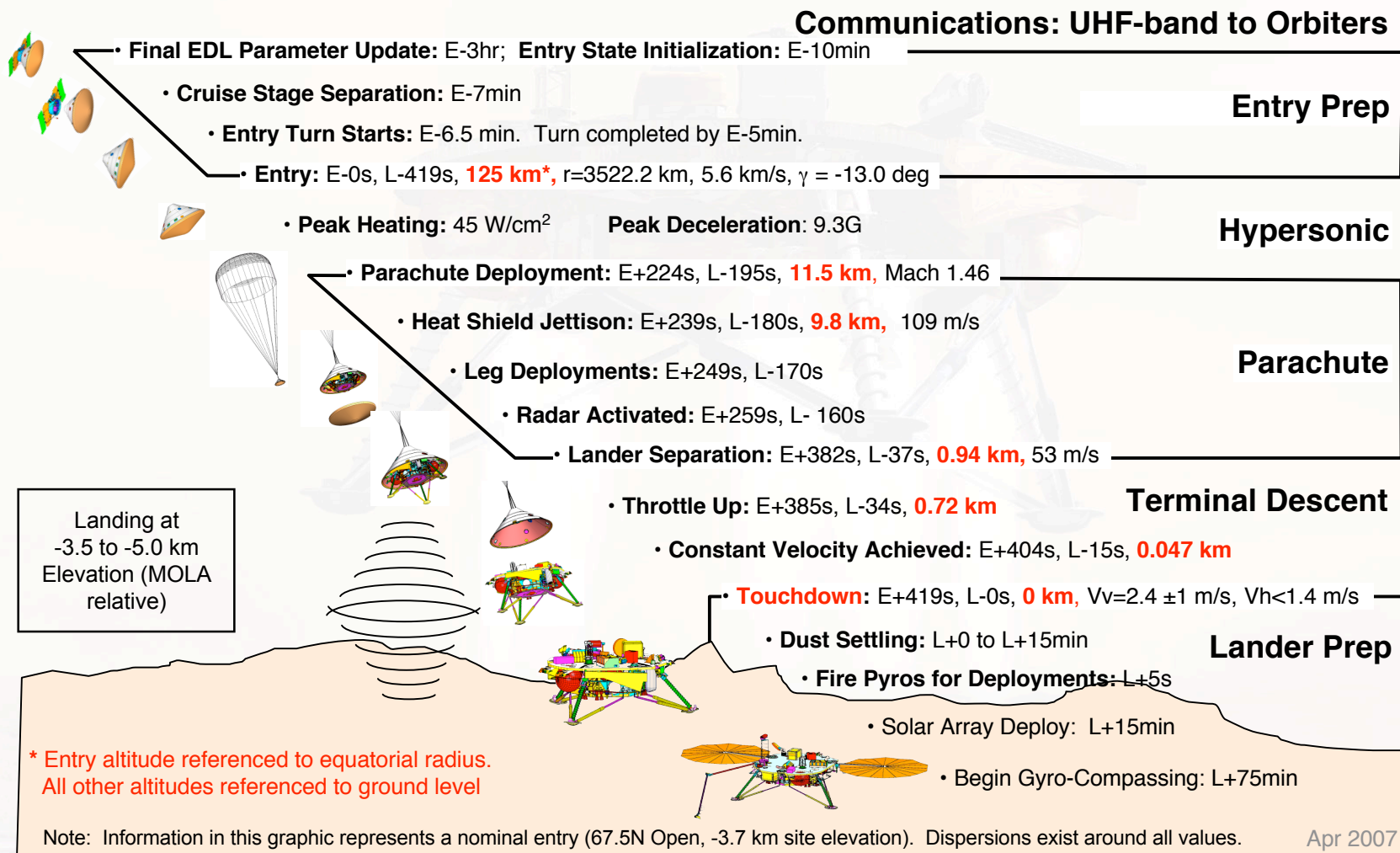


**Terminal Descent Configuration**





# Phoenix EDL Overview





# Mission Design Comparison



- **Mars 2001 Lander**
  - Equatorial landing region
  - 7.2 km/s entry velocity  $\Leftarrow$
  - +2.5 km (w.r.t MOLA) landing site elevation  $\Leftarrow$
- **Phoenix Lander**
  - Northern landing region: 65° N to 72° N
  - 5.8 km/s entry velocity  $\Leftarrow$
  - -3.5 km (w.r.t MOLA) landing site elevation  $\Leftarrow$



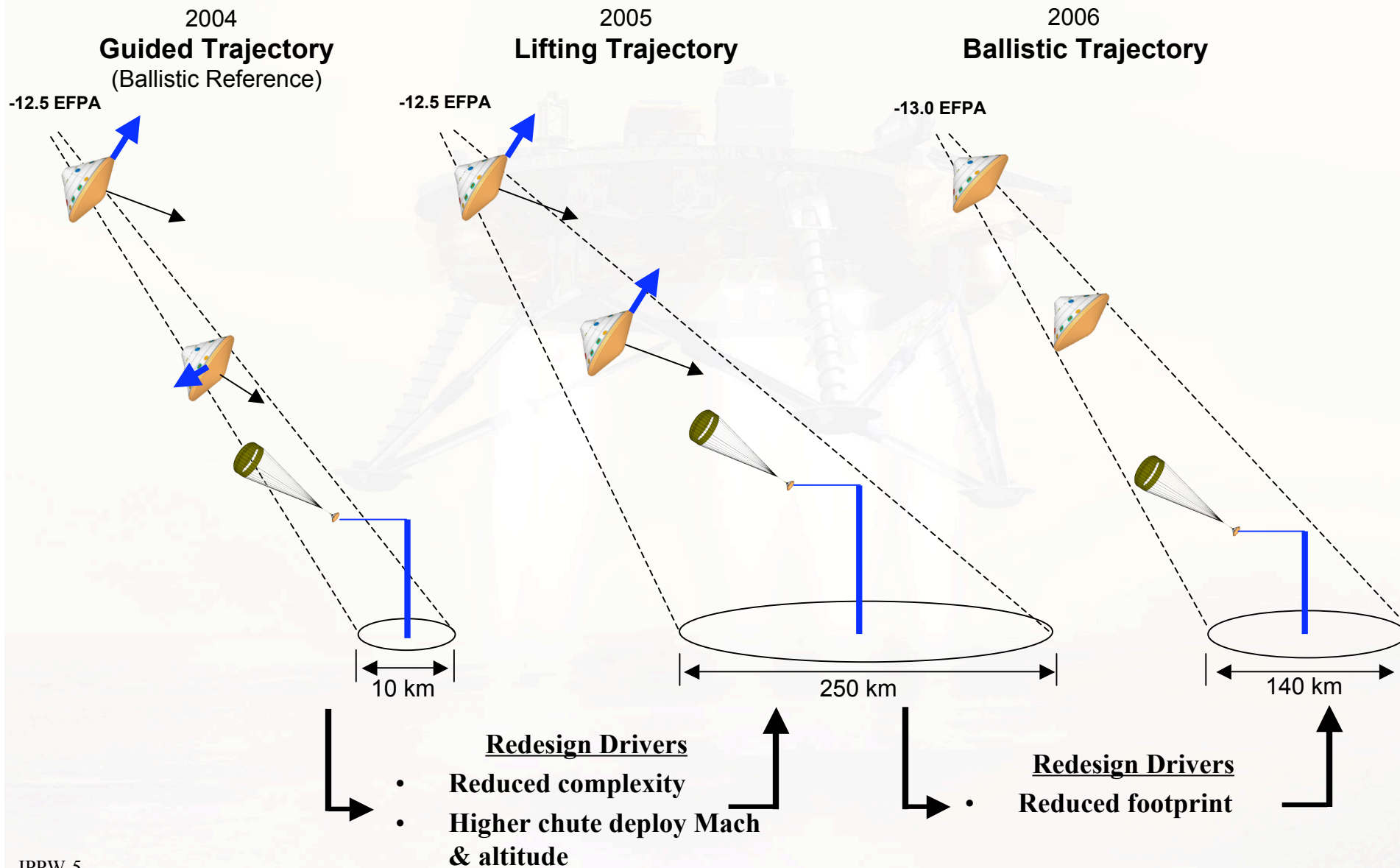
# Driving Constraints at Chute Deploy



- **Parachute Deploy Opening Loads**
  - Lander structure load limit is 12,700 lbs (51,893 N)
  - Constrains parachute opening dynamic pressure to 520 Pa
  - Results in EDL systems most rigid constraint
- **Parachute Deploy Mach Number**
  - Transonic instability forces need to keep Mach at chute deploy comfortably away from Mach 1.0
  - Angle of attach at chute deploy is critical for three reasons:
    - Parachute inflation qualification limits
    - Off-axial load limits on Lander structure
    - Vehicle oscillatory dynamics (wrist mode) while hanging from parachute
- **Parachute opening load drives down chute deploy point while Mach constraint drives up chute deploy point**



# Hypersonic Evolution (1/3)



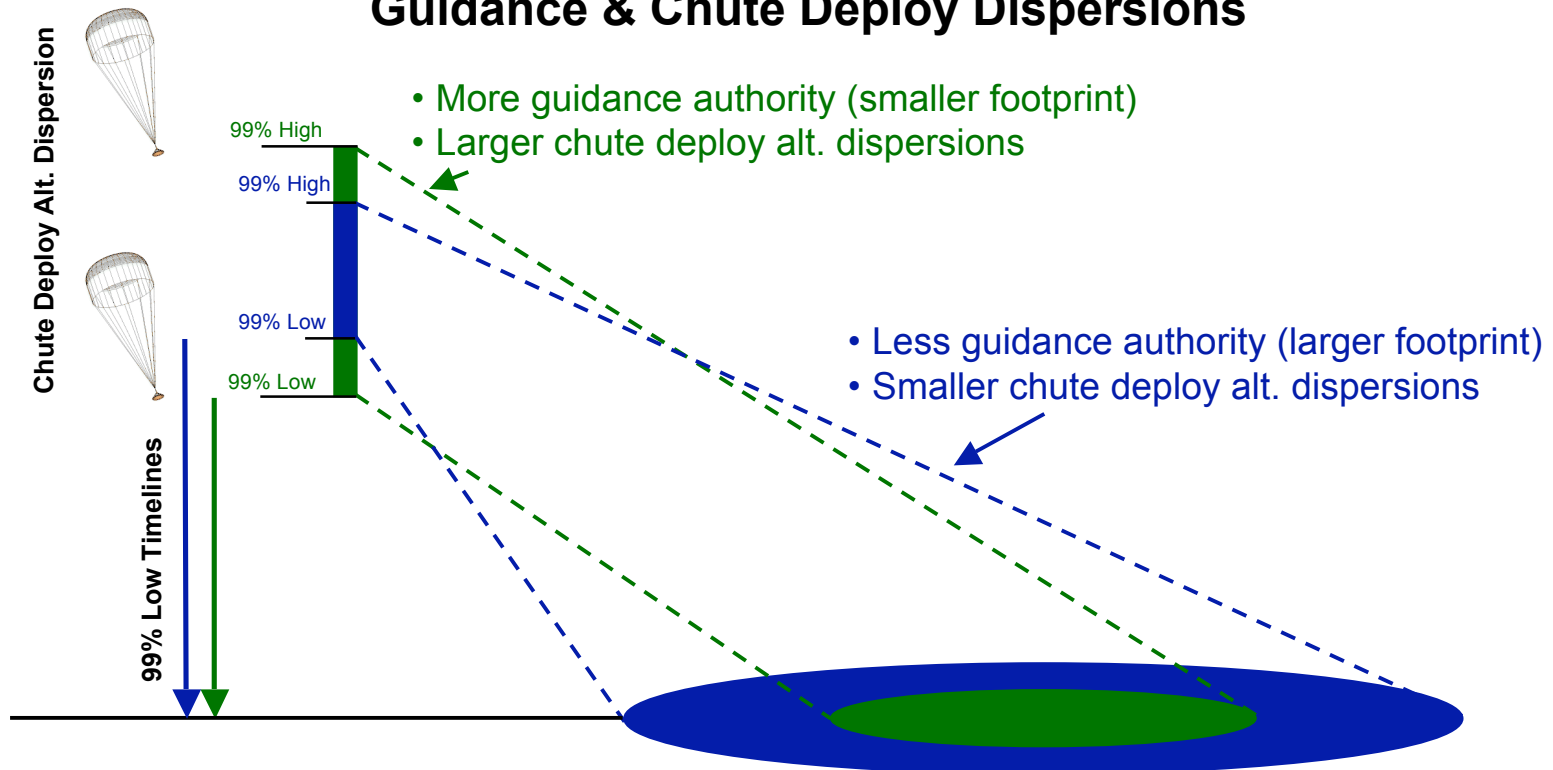




# Hypersonic Evolution (2/3)



## Guidance & Chute Deploy Dispersions

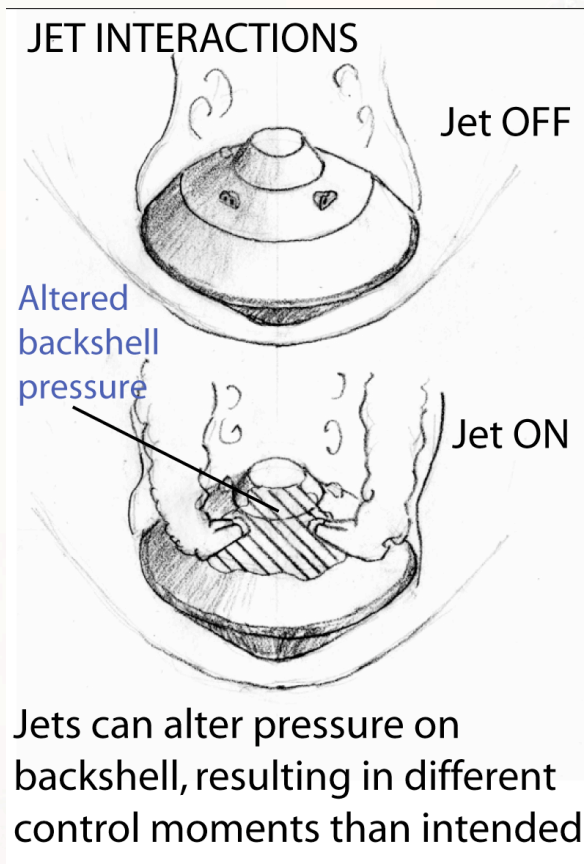




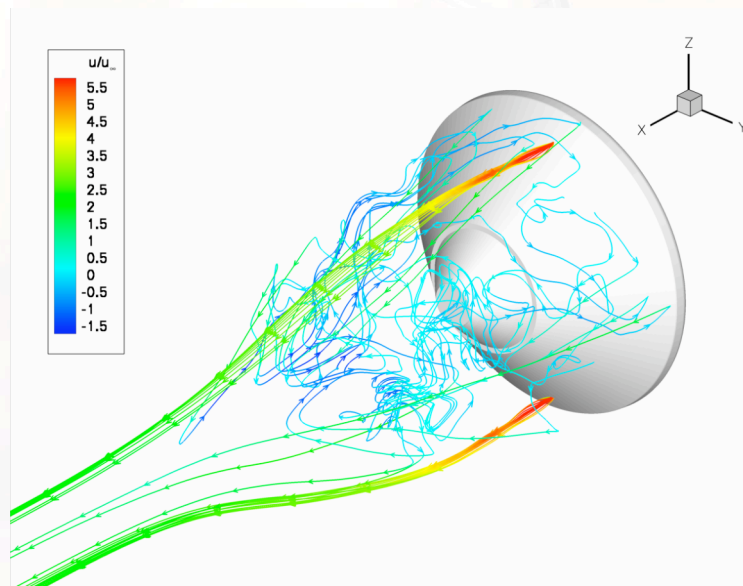
# Hypersonic Evolution (3/3)



## Aero/RCS Flow Interaction Phenomenon



- **CFD of Aero/RCS flow field shows potential for strong interaction from hypersonic regime to parachute deployment**
  - **RCS Pitch authority is degraded**
  - **RCS Yaw authority is low to non-existent (potential for control reversal exists)**
  - **Baseline is to increase control system deadbands to minimize/eliminate RCS thruster firings to avoid this flow interaction**



**CFD of Yaw  
Thruster Firing**



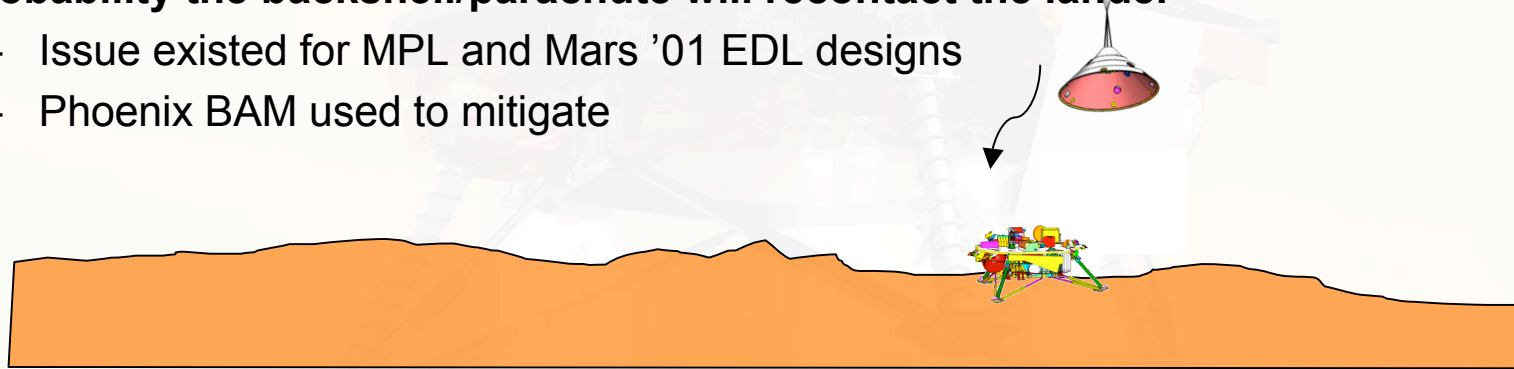
# Terminal Descent Evolution (1/2)



## Backshell Avoidance Maneuver (BAM)

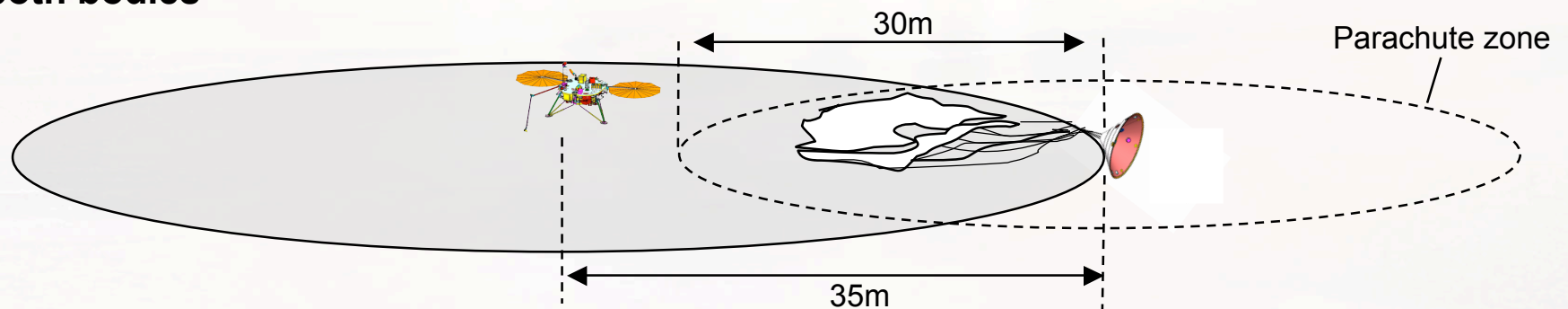
In cases of low wind and no wind terminal descent scenarios, there is an increased probability the backshell/parachute will recontact the lander

- Issue existed for MPL and Mars '01 EDL designs
- Phoenix BAM used to mitigate



### New Requirement

The distance between the center of mass of the lander and center of mass of the backshell shall be greater than 35m from 5s after lander separation to touchdown of both bodies





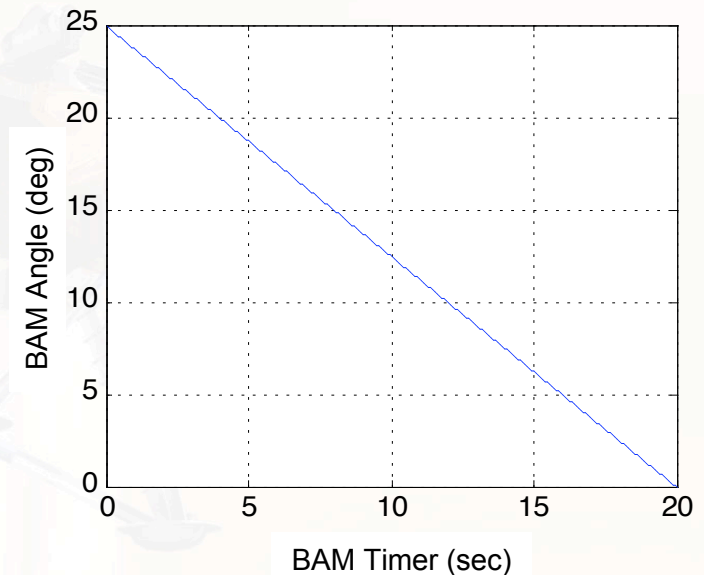
# Terminal Descent Evolution (2/2)



## BAM Design

Wind direction  
→

Extra delta-v in upwind direction



- Aim the maneuver upwind (Assumption: all residual horizontal velocity is due to wind)
- Upwind direction is estimated from the navigated horizontal velocity
- Start the BAM at tip up
- Compute the angle from horizontal velocity
- Two conditional tests constrain conditions of use



# Summary



- **Phoenix is a return to flight of the cancelled Mars '01 Lander, emphasizing thorough and extensive testing**
- **Mission design leads to more benign entry velocities and a much lower landing site elevation relieving pressure on EDL timeline**
- **Due to complexity, hypersonic guidance was de-scoped in favor of simple ballistic entry**
- **Immature understanding of thruster effectiveness in hypersonic/supersonic low led to relaxed attitude control**
- **A backshell avoidance maneuver was added to mitigate risk of backshell/parachute recontact of the Lander during terminal descent and at touchdown**
- **Changes to Phoenix EDL system architecture provides a more robust design for Mars EDL**